

Abstract

Stationary blade for a turbine and turbine

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The invention relates to a stationary blade (12) for a gas turbine (1) with a hollow sectional element (22) which extends radially with respect to the rotor (3) and has a platform (23) at each of its two ends, with a hollow inset (20) which is located in the sectional element (22) a certain distance from the inside (28) of the sectional element (22) and has a base (35) which faces one of the two platforms (23), with a coolant (K) flowing into the hollow space (21) of the inset (20) and flowing out through baffle cooling openings (29) provided on the inset (20) and with a recess (24) that is provided in the platform (23) located immediately opposite the base (35).

To specify a stationary blade (12) for which there is no mechanical damage during turbine operation it is proposed that the inset (20) extends into the recess (24) so that in the base area (30) of the inset (20) there are zones with reduced predefined flow rates.

FIG 2